# **TFAWS Interdisciplinary Paper Session**



# Proof of Concept Design and Analysis of Heat Reutilization of a Solid Oxide Electrolyzer Cell for Oxygen Supply

Samuel Ogletree, M.A. Rafe Biswas

Presented By Samuel Ogletree

ANALYSIS WORKSHOP

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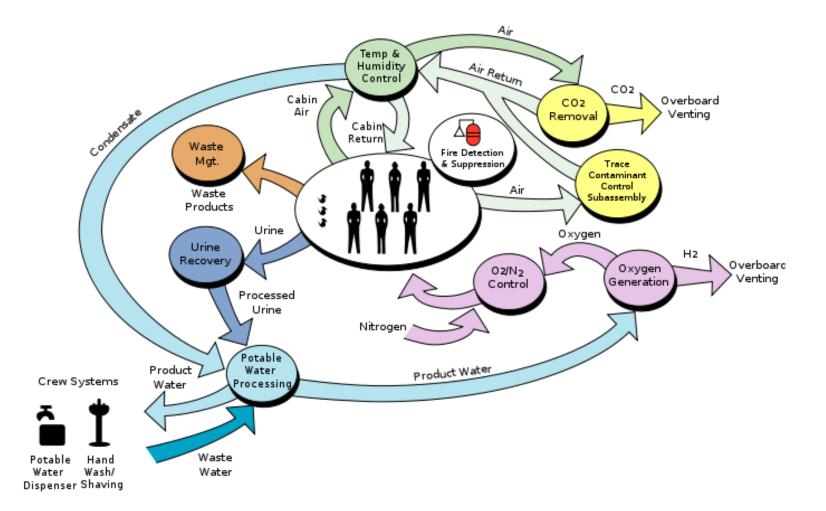


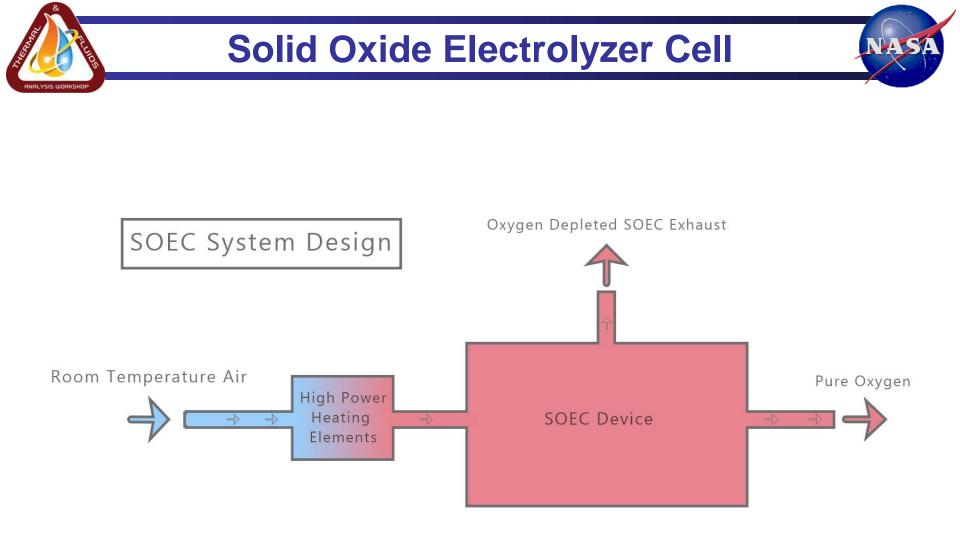
#### Innovation



# **Current Oxygen Production Systems**











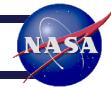
- Reliable large scale thermal management capabilities
  - High Temperature Operation
  - Minimize Power Consumption
- Scale in-situ oxygen production to more than 10 kg/day



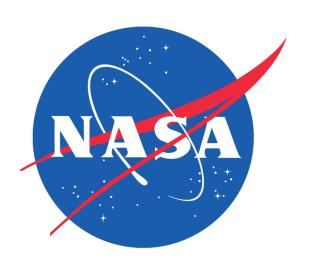




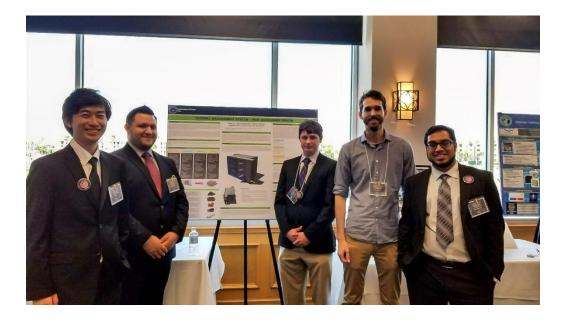
### **Student Partnership**

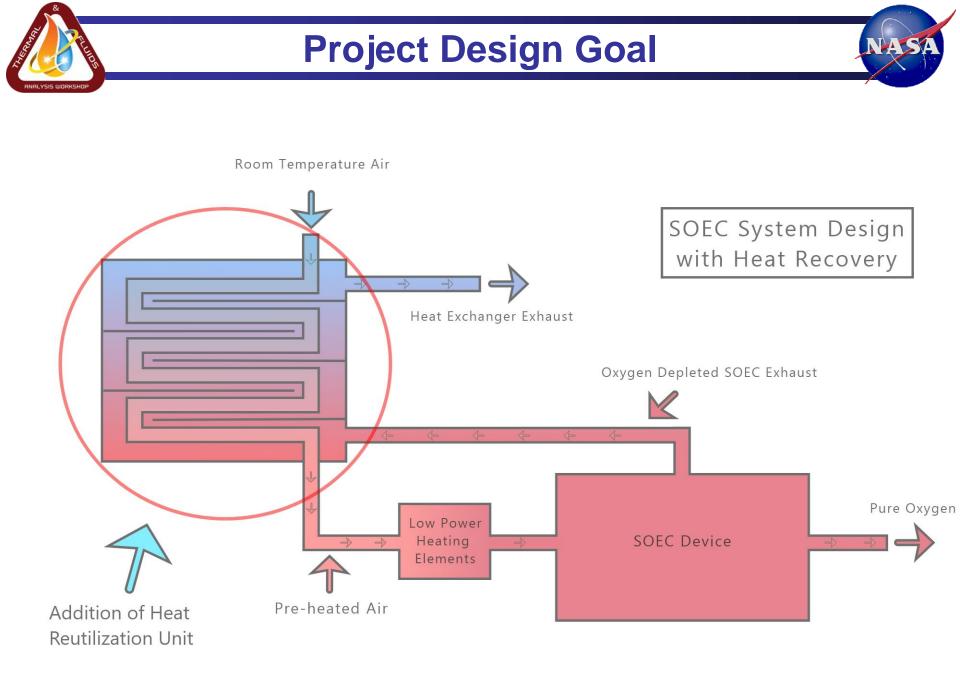






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Heat Exchanger Design Pugh Chart						
Desired Properties	Double-Pipe	Plate	Cross-Flow	Shell and Tube		
Pressure Drop	0	-	0	0		
HX size	-	+	+	0		
HT surface Area	0	+	+	+		
Cost	+	-	-	0		
Simplicity of Design	+	0	0	0		
Ease of Manufacturing	+	-	-	0		
Durability	0	+	0	0		
Maintenance	+	0	0	0		
Availability of Parts	+	-	-	+		
Effectiveness	-	+	-	+		
Design Flexibility	-	+	+	+		
Total	2	1	-1	4		
Rank	2	3	4	1		

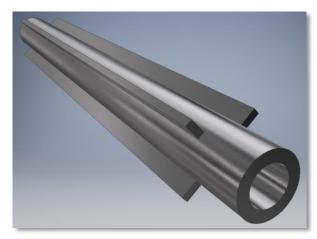


# **Final Design**













Shell		Tube	
Temperatures [°C]		Temperatures [°C]	
Inlet $(t_1)$ :	23.7	Inlet $(T_1)$ :	559.1
Outlet $(t_2)$ :	206.8	Outlet $(T_2)$ :	88.7
Differential ( $\Delta T_s$ ):	183.1	Differential $(\Delta T_t)$ :	470.4
Mass Flow Rate $[^{kg}/_{s}]$		Mass Flow Rate $[^{kg}/_{s}]$	
$2.99 \times 10^{-4}$		$1.95 \times 10^{-4}$	

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Analysis of Heat Exchanger:

- Reynolds Number
- Convective and Conductive Coefficients
- Overall Heat Transfer Coefficient
- Heat Duty
- Effectiveness
- Required Heater Power



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# **Assumptions and Properties**

- Steady-State Mass Flows
- Fully Developed Flows
- Constant Fluid Properties on shell and tube sides
- Flows Along respective Streamlines
- Well-insulated heat exchanger mass flows
- Constant heat flux along heat exchanger
- No external work
- Hot fluid in the tubes
- Counter current flow configuration

Fluid Properties				
<u>Symbol</u>	Property	Shell Fluid @ 115 °C	<u>Tube Fluid @ 324 °C</u>	
ρ	Density	$0.899 \frac{kg}{m^3}$	$0.586 \frac{kg}{m^3}$	
c <sub>p</sub>	Specific Heat	1.01 $kJ/kg \cdot K$	1.05 $^{kJ}/_{kg \cdot K}$	
ν	Kinematic Viscosity	$25.1 \times 10^{-6} m^2/_S$	$52.4 \times 10^{-6} m^2/s$	
k	Thermal Conductivity	$0.0324 \ ^{W}/_{m \cdot K}$	0.0455 <sup>W</sup> / <sub>m·K</sub>	







**Reynolds Number:** 

$$Re = \frac{VD}{v}$$

Where:

- Re is the Reynolds Number
- V is the average fluid velocity
- D is the flow section effective diameter
- v is the kinematic viscosity

Result:

Shell Flow:198.0[laminar flow]Tube Flow:127.6[laminar flow]







#### **Convective Coefficient:**

 $N_u k_f$ 

Where:

- h is the convective coefficient
- $N_u$  is the Nusselt Number
- $k_f$  is the fluid thermal conductivity
- D is the flow section diameter

#### Result:

Shell Convective Coefficient  $(h_o)$ : 1.82  ${}^{W}/{}_{m^2 \cdot K}$ Tube Convective Coefficient  $(h_i)$ : 12.5  ${}^{W}/{}_{m^2 \cdot K}$ 







Conductive Coefficient:

$$h_t = \frac{k_t}{L \cdot \ln(r_2/r_1)}$$

Where:

- $h_t$  is the conductive coefficient
- $r_1$  is the tube's inner diameter
- $r_2$  is the tube's outer diameter
- $k_t$  is the tube thermal conductivity
- *L* is the tube length

Result:

$$h_t = 63.9 \ ^{W}/_{m^2 \cdot K}$$





#### **Overall Heat Exchanger Coefficient:**

$$\left(\frac{1}{UA_{o}} = \frac{1}{h_{i}A_{i}} + \frac{1}{h_{t}A_{o}} + \frac{1}{h_{o}A_{o}}\right)$$

#### Where:

- U is the overall heat transfer coefficient
- $A_i$  is the tubes inner surface area
- $A_o$  is the tubes outer surface area

#### Result:

# Overall Heat Transfer Coefficient: 1.48 $W/_{m^2 \cdot K}$







$$Q = UAT_{LMTD}$$

Where:

- Q is the heat duty
- A is the heat transfer surface area
- $T_{LMTD}$  is the log mean temperature difference

Result:

# Heat Duty: 70.7W







HX Effectiveness:

$$\varepsilon = \frac{t_2 - t_1}{T_1 - t_1}$$

Where:

- ε is effectiveness
- $t_1$  is shell inlet temperature
- $t_2$  is shell outlet temperature
- $T_1$  is tube inlet temperature

Result:

#### Effectiveness: 0.342





$$\left[E = \dot{m}c_p \Delta T\right]$$

Where:

- E is the required power
- $\dot{m}$  is the mass flow rate
- $c_p$  is initial temperature heat capacity
- $\Delta T$  is the needed change in temperature

#### **Result:**

Power Required			
Without Heat Reutilization	With Heat Reutilization		
303.8 <i>W</i>	232.1 W		





- The biggest resistances to heat transfer are shell and tube fluid convection coefficients
  - $\circ~$  Low flow velocity in both channels
- Tube structure had minimal affect on overall heat transfer resistance
- 25% reduction in heater power is a great start, but further modifications can be made to increase power reduction.



### **Future Considerations**

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### Modified Tubing Design



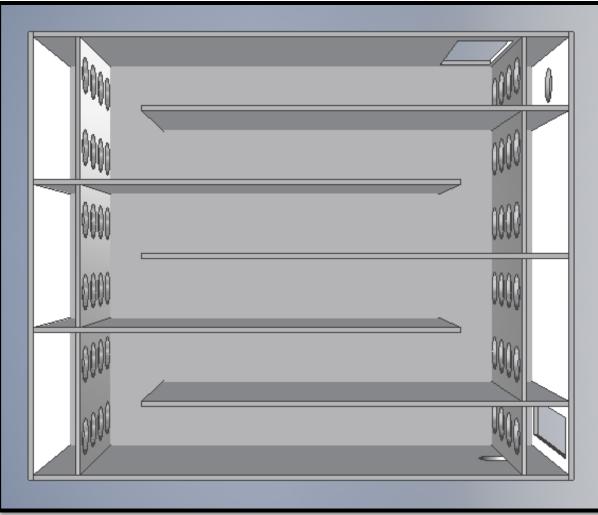






### **Future Considerations**

#### Modified Flow Design



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- Texas Space Grant Consortium
- University of Texas at Tyler advisor Dr. Andres Garcia
- NASA mentor Dr. John Graf
- Houston Custom Metal Works
- KBRWiley



#### References



[1] Barry, Patrick. "*Breathing Easy on the Space Station.*" NASA. [online]. Website: https://science.nasa.gov/science-news/science-at-nasa/2000/ast13nov\_1

[2] Sridhar, K.R. "Oxygen Production on Mars Using Solid Oxide Electrolysis." Solid State Ionics. vol. 93. pp 321-328.

[3] Janna, William. *Design of Fluid Thermal Systems*. Delhi, India: Cengage Learning, 2015.

[4] F. P. Incropera, D. P. Dewitt, T. L. Bergman, A. S. Lavine. *Principles of Heat and Mass Transfer.* Delhi, India: Wiley, 2016.

[5] Texas Space Grant Consortium, "A Low Power, Solid State, Method of Oxygen Supply." 12 9 2017. [Online]. Available:

http://www.tsgc.utexas.edu/challenge/PDF/topics/Topic\_TDC\_37\_F17.pdf.

[6] Texas Space Grant Consortium. "*TSGC Design Challenge.*" TSGC. [online]. Website: http://www.tsgc.utexas.edu/challenge/

[7] Wischnewski, Berndt. "*Calculation of Thermodynamic State Variables of Air*". Peace Software. [online]. Website: http://www.peacesoftware.de/einigewerte/luft\_e.html









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